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(54) **Blade with platform cooling**

(57) A cooled turbine blade for a gas turbine engine has a platform cooling supply channel 34 that supplies cooling air to a plurality of cooling holes 36 in the surface

28 of the blade platform 24 adjacent the trailing edge 22 of the blade. The passage 34 extends between the sides 30, 32 of the blade platform 24 and communicates with feather seal slots 42, 44 provided on the blade platform.

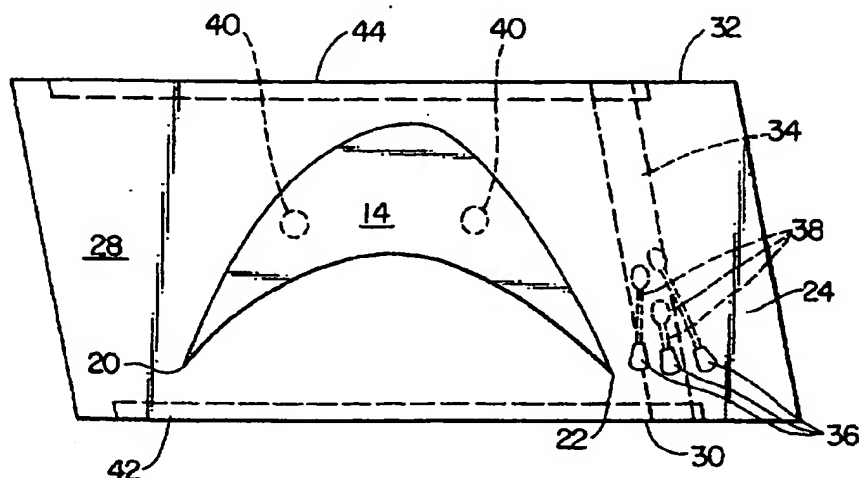


FIG. 2



European Patent
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EUROPEAN SEARCH REPORT

Application Number
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Category	Citation of document with indication, where appropriate, of relevant passages	Relevant to claim	CLASSIFICATION OF THE APPLICATION (Int.Cl.7)
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The present search report has been drawn up for all claims			
Place of search MUNICH		Date of completion of the search 31 October 2001	Examiner Koch, R
<p>CATEGORY OF CITED DOCUMENTS</p> <p>X : particularly relevant if taken alone Y : particularly relevant if combined with another document of the same category A : technological background O : non-written disclosure P : intermediate document</p> <p>T : theory or principle underlying the invention E : earlier patent document, but published on, or after the filing date D : document cited in the application L : document cited for other reasons % : member of the same patent family, corresponding document</p>			

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**ANNEX TO THE EUROPEAN SEARCH REPORT
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